Problem %

Consider the following second-order system with an extra pole:

$$H(s) = \frac{\omega_n^2 p}{(s+p)(s^2 + 2\zeta\omega_n s + \omega_n^2)}.$$

Show that the unit step response is

$$y(t) = 1 + Ae^{-pt} + Be^{-\sigma t}\sin(\omega_d t - \theta),$$

where

$$\begin{split} A &= \frac{-\omega_n^2}{\omega_n^2 - 2\zeta\omega_n p + p^2}, \\ B &= \frac{p}{\sqrt{(p^2 - 2\zeta\omega_n p + \omega_n^2)(1 - \zeta^2)}}, \\ \theta &= \tan^{-1}\frac{\sqrt{1 - \zeta^2}}{-\zeta} + \tan^{-1}\frac{\omega_n\sqrt{1 - \zeta^2}}{p - \zeta\omega_n}. \end{split}$$

- (a) Which term dominates y(t) as p gets large?
- (b) Give approximate values for A and B for small values of p.
- (c) Which term dominates as *p* gets small? (Small with respect to what?)
- (d) Using the preceding explicit expression for y(t) or the step command in MATLAB, and assuming that $\omega_n = 1$ and $\zeta = 0.7$, plot the step response of the preceding system for several values of p ranging from very small to very large. At what point does the extra pole cease to have much effect on the system response?

Problem &

Suppose that unity feedback is to be applied around the listed open-loop systems. Use Routh's stability criterion to determine whether the resulting closed-loop systems will be stable.

(a)
$$KG(s) = \frac{4(s+2)}{s(s^3+2s^2+3s+4)}$$

(b)
$$KG(s) = \frac{2(s+4)}{s^2(s+1)}$$

(c)
$$KG(s) = \frac{4(s^3+2s^2+s+1)}{s^2(s^3+2s^2-s-1)}$$

Problem 3

Use Routh's stability criterion to determine how many roots with positive real parts the following equations have:

Problem (

Find the range of K for which all the roots of the following polynomial are in the LHP:

$$s^5 + 5s^4 + 10s^3 + 10s^2 + 5s + K = 0.$$

Use MATLAB to verify your answer by plotting the roots of the polynomial in the s-plane for various values of K.

Problem 5

Steady State Error Analysis for a P-Controlled System



Consider the P-controlled system in the above figure.

a) Express the output y(s) as a function of the command input r(s) and the disturbance d(s).

b) Express the following error e(s) as a function of the command input r(s) and the disturbance d(s).

c) Determine the steady-state value of the following error $e_{ss} = \lim_{t \to 0} e(t)$ when the command is a ramp input $r(s) = A/s^2$, and the disturbance is a step d(s) = D/s. (Use the Final Value Theorem).

d) Discuss how the natural frequency ω_n and damping ratio ζ of the closed loop poles affects the steady state error.